

# AERODYNAMIC PERFORMANCE OF AN ULTRA-LOW ASPECT RATIO CENTRIPETAL TURBINE STATOR

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## ABSTRACT

An extensive measurement campaign was carried out at the von Karman Institute for Fluid Dynamics to assess the aerodynamic performance of an ultra-low aspect ratio centripetal turbine stator. The test section consisted of 18 periodic sectors of two blade passages each, tested at transonic conditions in a blow-down facility. Particle image velocimetry (2D-PIV) measurements were performed in the trailing edge area at blade mid-height. Further downstream, 16 micro virtual 3-hole pressure probes were used to measure the aerodynamic performance at about 44% of the radial chord, downstream of the trailing edge. Results describe a highly swirling flow-field characterized by a very large tangential velocity component increasing at smaller radii. The radial component decreased instead to compensate for an opening of the test section closer to its center, enhancing therefore the tangential nature of the generated flow field. Adjacent blades showed asymmetric wakes as an effect of the particular design of the nozzle. The performance analysis exhibited very high loss coefficients (always higher than 12%) which have to be related to the extremely small aspect ratio of the cascade.

## NOMENCLATURE

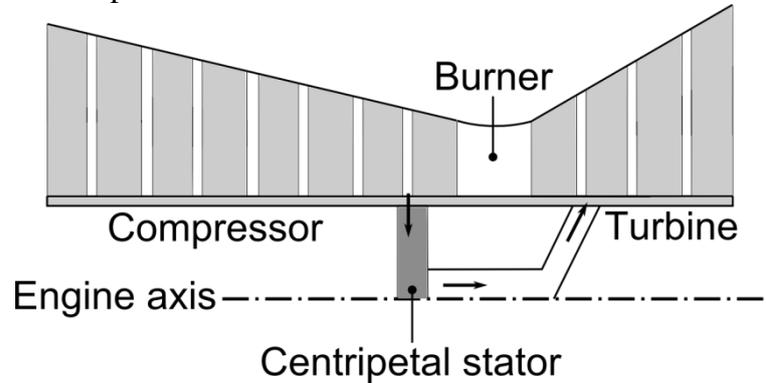
$C_r$	Radial chord [mm]	<b>Subscripts</b>	
$d$	Diameter [mm]	0	Total conditions
Ma	Mach number [-]	1	Inlet section
Re	Reynolds number [-]	2	Outlet section
P	Pressure [Pa]	rad	Radial component
R	Radius [mm]	tg	Tangential component
T	Temperature [K]		
V	Velocity [m/s]	<b>Acronyms</b>	
		PIV	Particle Image Velocimetry
		VKI	von Karman Institute for Fluid Dynamics
<b>Greeks</b>		LE	Leading edge
$\beta$	Flow angle [°]	TE	Trailing edge
$\delta$	Measurement uncertainty [-]		
$\vartheta$	Angular coordinate [°]		
$\zeta$	Kinetic energy loss coefficient [-]		

## INTRODUCTION

In the last decades, the typical design targets for new aero-engine architectures always involved an increase of the efficiency and of the power output of the thermodynamic cycle by maximizing the turbine inlet temperature. As a consequence, the heat load to the high pressure turbine (HPT) rose as well, forcing the use of a colder cooling flow in order to meet the required thermal protection needs. A reduction of the coolant temperature can be obtained by means of an expansion along the coolant delivery circuit, before feeding the blade and the disk area. Consequently, after flowing over the disk surfaces, the cooling air is then injected into the mainstream from the inter-platform gap where it is further used to protect the leading edge region of the endwall. In the latter

application, in particular, a high compatibility with the flow angles of the main stream is also recommended to reduce as much as possible the aerodynamic impact of the injection.

A possible design which meets all these requirements consists in a centripetal turbine stator located in the core of the engine, at a smaller radius than the endwall (see Figure 1). This configuration guarantees moreover a high degree of compactness of the component, leading anyway to the adoption of ultra-low aspect ratio blades.



**Figure 1: typical engine arrangement of a centripetal turbine stator.**

The development of such a component faces a deep lack of publications in the open literature. The most complete contribution on this topic available today was provided by Simpson et al. (2013) who performed an extensive numerical study describing the effect of solidity and vaneless space on the performance of a low aspect ratio automotive radial turbine stator. This experimental activity was limited to the measurement of the downstream wall static pressure distribution which was then used to evaluate the quality of the numerical predictions. Results were given mainly in terms of mass-averaged kinetic energy loss coefficient ( $\zeta$ ) and showed that the solidity had the strongest effect on the stator efficiency:  $\zeta$  grew almost linearly with the number of blades of the cascade, leading to values as high as 20% for the 30 blades stator configuration.

An extensive measurement campaign was recently carried out at the VKI on a transonic ultra-low aspect ratio centripetal turbine stator with the intent of providing a detailed insight into the phenomenology of the flow field that such a component generates. Total pressure, velocity and flow angle distributions were measured at two different radial locations downstream of the nozzle. Near to the trailing edge, the performance of the cascade was assessed by particle image velocimetry (PIV) measurements which were then processed with an advanced PIV processor. Further downstream, the total pressure and flow angle distributions were measured by means of 16 micro virtual 3-hole probes.

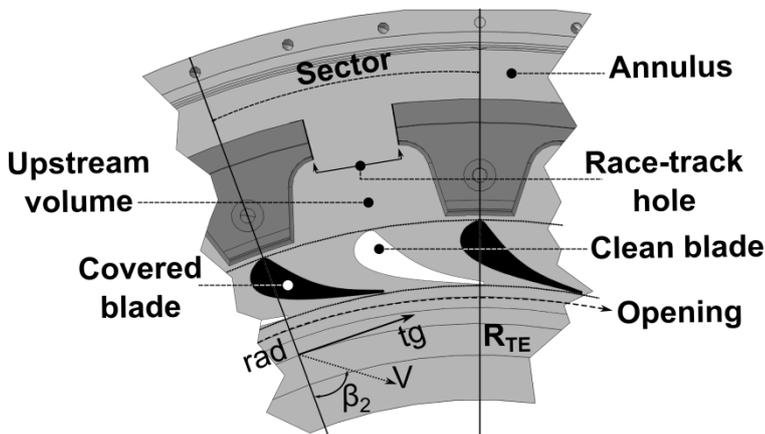
In the present contribution, the test section is described and the measurement techniques discussed, paying a particular attention to the issues related to the challenging geometrical constraints that directly drove and affected the design of the experiments. Results are successively presented and critically analyzed in terms of total pressure wakes, flow angle and loss coefficient distributions at two different radial positions in the discharge section of the cascade.

## **TEST SECTION, INSTRUMENTATION AND TEST CONDITIONS**

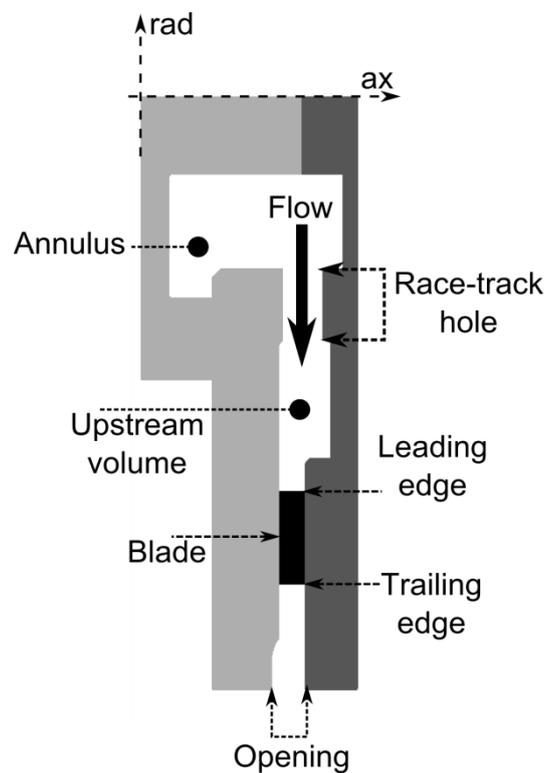
The test section consisted of 18 periodic sectors 20 [°] wide. As visible in Figure 2 (schematic view not on scale of the test section), each sector consisted of a “clean” and a “covered” blade. The “clean” blade had the leading edge aligned with the mean line of the upstream feeding hole while the inlet section of the “covered” one was instead blocked by a wall for fixing purposes. According to the principle sketch of the meridional flow path presented in Figure 3, downstream the race-track hole a settling volume was placed which fed the bladed part of the test section. The latter was only 6.8 [mm] high to become 8.4 [mm] further downstream of the trailing edge radius, where an opening was present. The facility is a typical blow-down wind tunnel: the flow was drawn from a pressurized reservoir and radially delivered to the upstream distribution annulus at temperatures

ranging between 10 and 15 deg.. After flowing through the cascade, it was then discharged axially at atmospheric conditions. Tests were carried out in full Mach number ( $Ma$ ) similarity with engine conditions while the effect of the lower total inlet temperature on the Reynolds number ( $Re$ ) was partially counter-balanced by the use of a scaled model which allowed to retrieve  $Re$  values of the same order of magnitude as the engine ones, sufficiently high to consider a turbulent flow

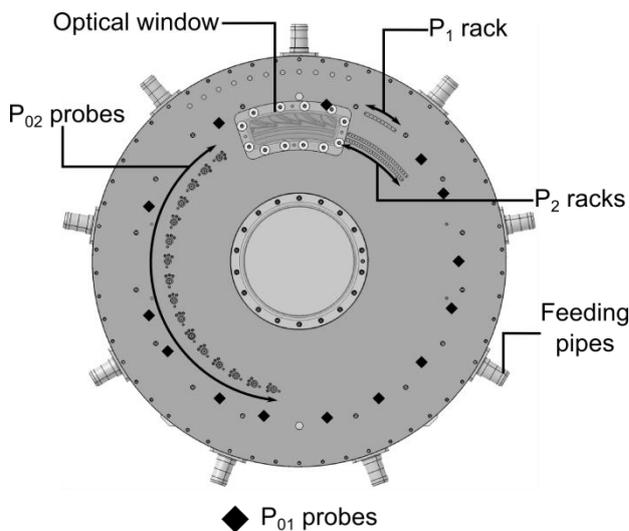
The test section was sealed from the exterior by means of a carter which was directly carrying “on-board” the optical access window for PIV measurements as well as all the rest of the instrumentation, as shown in Figure 4. The homogeneity of the total inlet pressure  $P_{01}$  was checked by means of 10 total pressure probes located at 31.7% of the radial chord ( $C_r$ ) upstream of the leading edge plane and distributed over many different sectors. At the same radial location, one sector was equipped with a rack of 9 static pressure taps to retrieve the azimuthal wall static pressure distribution  $P_1$ . The downstream wall static pressure  $P_2$  was measured by means of two rows of 25 static pressure taps located respectively at 8.6% $C_r$  and 44.5% $C_r$  downstream of the trailing edge, covering 1.5 sectors in total. The downstream total pressure ( $P_{02}$ ) distribution was also measured at 44.5% $C_r$  by 16 micro cylindrical pressure probes operated in a virtual 3-hole mode. The probes were distributed circumferentially over several sectors in such a way that they covered an entire sector (i.e. 2 adjacent blade passages). Tests were carried out at nearly ambient inlet total temperature and at a target isentropic discharge Mach number ( $Ma_{2s}$ ) of 0.9.



**Figure 2: frontal view of a single sector of the test section: adopted conventions and nomenclature.**



**Figure 3: principle sketch of the meridional flow path (not to scale).**



**Figure 4: view of the instrumented carter.**

# EXPERIMENTAL METHODOLOGY

## Particle image velocimetry

The PIV measurement setup is shown in Figure 5. The laser beam, provided by a Quantel pulsed Twin CFR Nd:YAG laser (532 [nm]) light source, was turned into the direction of the test section axis by a right angle prism. A cylindrical lens was used to open the beam into a laser sheet which was then focused by a 500 [mm] focal length lens. A second right angle prism was then fitted into the facility to turn the laser sheet parallel to the cascade endwall. The resulting PIV measurement plane was set at the blade's mid-height; the light sheet thickness was about 1 [mm] in the trailing edge region. PIV images were recorded by means of a PCO 1.3 megapixel Sensycam CCD camera, fitted with a 50 [mm] NIKKOR objective. The limited access provided by the optical Plexiglass window led to a PIV interrogation area restricted to the suction side of a covered blade, its corresponding throat region and part of the downstream section, as shown in Figure 6.

Since the tests were carried out at transonic conditions, a separation time of 0.78 [ $\mu$ s] was adopted. Incense smoke was used as tracer as it contains particles of about 1 [ $\mu$ m] in size (Dupont et al., 2008). The smoke was provided from a pressurized vessel, equipped with two hotplates used to burn commercially available granulated incense. The injection point was located upstream the leading edge plane, in the "trapezoidal" volume, where a total pressure probe was also present to verify the homogeneity of the inlet conditions with respect to the other sectors.

During the normal test time lapse, the temperature of the wetted surfaces of the facility drastically reduced because of the total inlet temperature drop (typical of every blow-down facility) and of the flow static temperature reduction imposed by the expansion. In these conditions, the small fraction of water vapor suspended in the flow close to the endwalls condensed and generated a variation of the reflections pattern in time. Moreover, the thin condensed layer was acting as a trap for the small incense particles, leading to the entrainment of some of them by the endwall roughness and therefore to the generation of stationary bright spots in the background. Given the extremely short distance between the PIV measurement plane and the endwall (about 3 [mm]), the stationary spots were anyway recorded by the camera and directly generated a zero displacement vector also at the smallest aperture. In order to get rid of this type of artifacts, images were pre-processed by means of an in-house code that was performing a pixel-wise minimum intensity subtraction on each image pair individually. Stationary particles and variable reflections effects could be therefore efficiently removed.

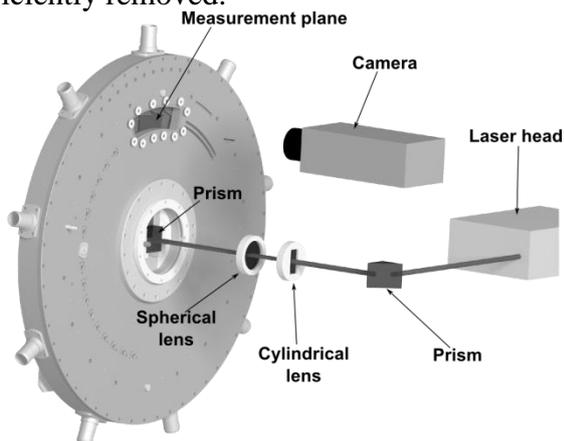


Figure 5: PIV optical setup arrangement.

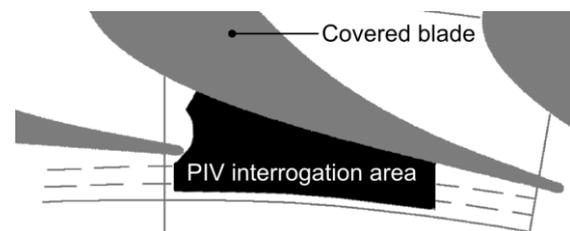


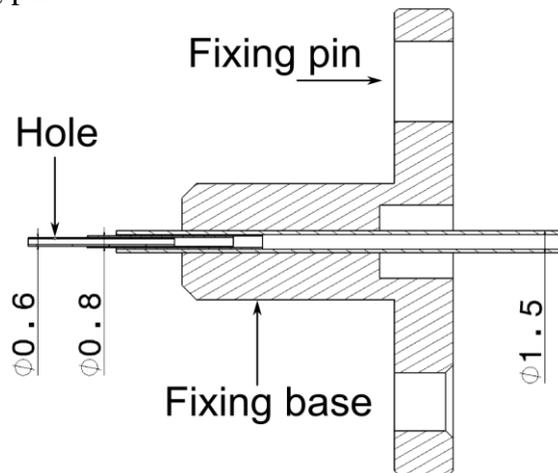
Figure 6: field of view of PIV measurements.

The PIV measurements were performed in a region characterized by a high vorticity level and strong velocity gradients. In these conditions, standard commercial PIV codes demonstrated to be inefficient as shown by Fontaneto et al. (2014). In order therefore to retrieve reliable velocity fields, the advanced adaptive PIV processor ARPIV developed by Theunissen (2010) was used. For each image pair, the code first performs a particle recognition analysis. Interrogation windows are then

displaced all over the field of view, providing to maintain a minimum number of particles as set by the operator. The code refines, through successive iterations, the position and the size of the interrogation cells, adapting them according to the displacement vector computed at the previous step but also maintaining a high number of particles for each cell.

### Micro virtual 3-hole probes

A set of 16 micro cylindrical probes was placed downstream of the trailing edge, at  $44.5\%C_T$ . The probes were distributed over several sectors (see Figure 4) in order to reduce as much as possible the blockage effect but in such a way to cover a full sector periodicity with an angular step of  $1.25 [^\circ]$ . As shown in Figure 7, the measuring head of the probe, the only part exposed to the flow, consisted of a  $0.6 [mm]$  external diameter pipe with a  $0.1 [mm]$  drilled pressure tap. This head was welded to a fixing base which guaranteed a univocal positioning from the exterior of the facility by means of a fixing pin.



**Figure 7: cut-view of the micro virtual 3-hole pressure probes.**

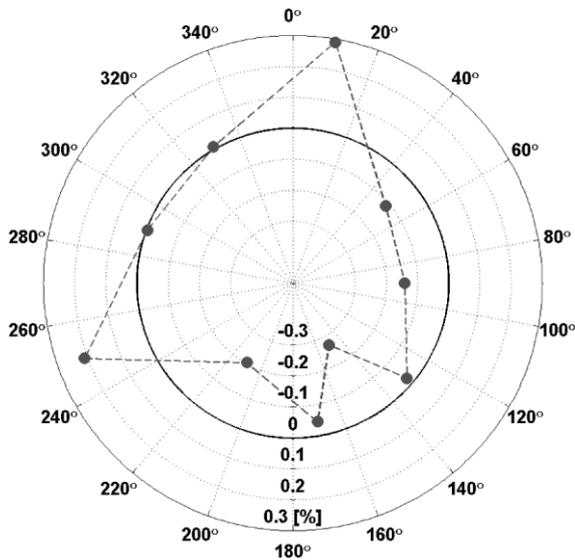
The probes were operated in a virtual 3-hole mode, allowing to retrieve the downstream flow angle ( $\beta_2$ ) and total pressure distributions. This was performed by rotating the probes in three different positions with an angular step of  $35 [^\circ]$  and by applying an angular calibration previously performed. Even though it was possible to modify the angular position of the probes without opening the test section, the facility had anyway to be switched off inducing a slight variation of the upstream total pressure between the three different angular positioning. In order therefore to enhance the quality of the results, all pressure measurements were normalized with respect to a reference inlet total pressure value prior to reducing the data. The complete procedure is described by Fontaneto et al. (2014).

Given the very small dimensions of the pressure tap, the dynamic response of probes was checked with the intent of defining the correct settling time to be due before the start of the acquisition. The transfer function of such a pneumatic system was evaluated by means of the model of Bergh and Tijdeman (1965), considering typical upstream total conditions. The result showed a classical first order system behavior whose time constant was found to be about  $0.2 [s]$  at the 3dB cutoff frequency. Since the used model necessarily introduces some degrees of approximation, it was in the end decided to extend it to  $2 [s]$ .

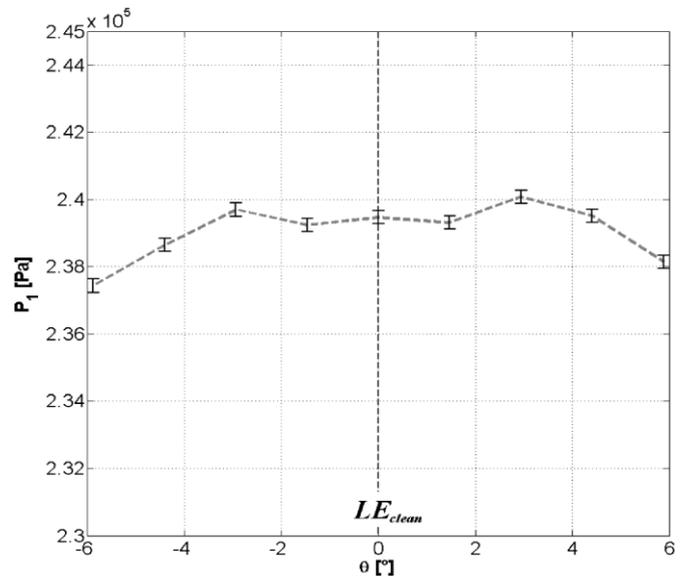
## INLET HOMOGENEITY AND OUTLET PERIODICITY

As mentioned previously, the testing air was provided to every sector by means of a large-volume feeding annulus which guaranteed a high degree of uniformity of the inlet conditions. This is confirmed by the polar plot of Figure 8 where the relative difference between the local measured total pressures and the global mean value never exceeds  $\pm 0.3\%$ . The  $P_{01}$  distribution was measured by means of 10 probes displaced over as many sectors and aligned with the LE of the “clean” blade.

At the same radial coordinate, 9 static pressure taps were then distributed along the inlet trapezoidal cavity of a typical sector (see Figure 4). The resulting P1 distribution is shown in Figure 9 and it is characterized by a flat evolution in the central region (centered with the “clean” blade) which drops by about 1% at the sides, close to the “covered” blades inlet areas.

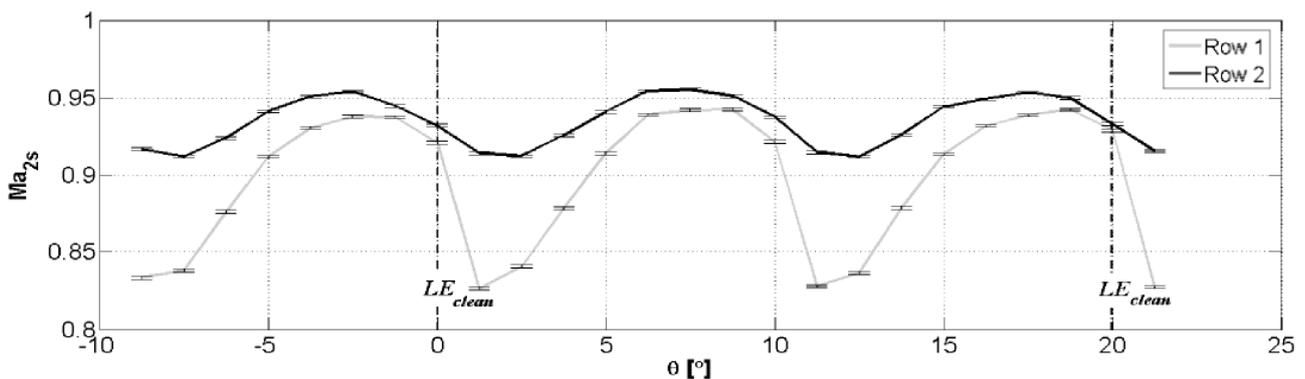


**Figure 8: polar distribution of the relative difference of  $P_{01}$  local value with respect to the mean**



**Figure 9: inlet static pressure distribution in the trapezoidal upstream volume**

The periodicity of the flow in the discharge section was verified by means of two rows of 25 static pressure taps each and respectively located at 8.6%  $C_r$  and 44.5%  $C_r$  downstream of the trailing edge. The pitch between the taps was such that 1.5 sectors could be measured. The corresponding isentropic discharge Mach number ( $Ma_{2s}$ ) is plotted in Figure 10 against the azimuthal coordinate and describes a marked free vortex behavior of the flow field generated in this region: while flowing towards the outlet channel, at the center of the test section, the flow is further accelerated, reaching an almost sonic condition on row 2. While flowing from row 1 to 2, the distribution becomes more symmetrical and homogeneous even though a high degree of periodicity could always be recognized, with no evident differences between “covered” and “clean” blades.



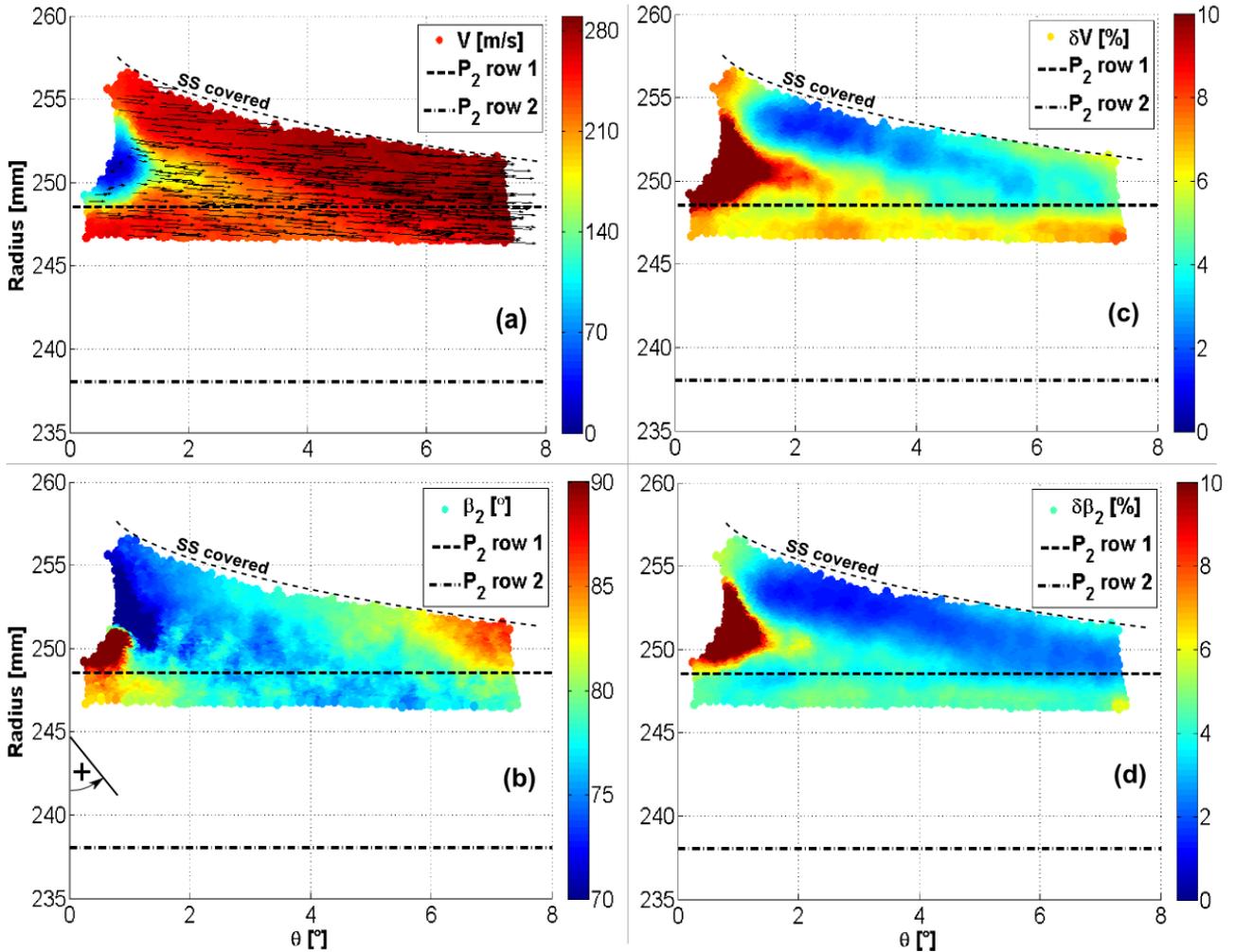
**Figure 10: isentropic downstream Mach number distribution on row 1 and row 2.**

## PIV results

The contour plots shown in Figure 11 report the velocity field (a) and the flow angle distribution (b) measured in the frame of the PIV campaign. The quantities are plotted with respect to the local polar coordinates;  $R$  is the radial distance from the center of the test section and  $\vartheta$  the azimuthal

shift from the leading edge of a “covered” blade located in 0. On the right hand side of the figure, the computed uncertainties are provided as a percentage of the local measured value.

In general, the flow field is characterized by a high velocity level, specifically in the throat region, all along the suction side of the covered blade and at the bottom, at smaller radii. From the trailing edge of the adjacent vane, the velocity deficit in the wake extends till  $\vartheta=4$  [°] where it is then mixed out by the upper and the lower high momentum flow regions. The velocity increases to values as high as 270 [m/s]. From  $\vartheta=5$  [°] till the end of the field of view, at  $\vartheta=7.3$  [°], the trace of the wake cannot be appreciated anymore, with high velocities at every radial coordinate. The latter part of the flow field directly feeds the next sector, generating therefore the high momentum region under the wake at smaller radii.



**Figure 11: PIV measurement results; (a) absolute velocity, (b) flow angle, (c) absolute velocity and (d) flow angle uncertainties. Velocity vectors were under-sampled by a factor 10.**

The absolute velocity contour plot described up to this point is necessarily compatible with a strong swirled behavior of the flow field as confirmed by the flow angle distribution whose magnitude remains higher than 75 [°] almost everywhere in the investigated area. In any case,  $\beta_2$  is always smaller than  $\pi/2$ , meaning that the stream tubes would show the tendency to reduce their radial coordinate while flowing clockwise. By reducing the radial coordinate, the absolute velocity is necessarily boosted by the conservation of the angular momentum, imposing therefore a further acceleration which justifies the higher  $Ma_{2s}$  measured on the second row of static taps than on row 1 (see Figure 10).

Measurement uncertainties were computed by taking into account the local standard deviation ( $\sigma_x$ ) and the number of recorded images ( $N_{records}$ ), as described by Equation 1;  $Z_{\alpha}$  is the 95<sup>th</sup>

percentile of the standard normal distribution (95% confidence interval). Both  $\delta V$  and  $\delta\beta_2$ , shown in Figure 11c and Figure 11d respectively, present the highest uncertainties in the wake region of the “clean” blade. In fact, under the effect of the centrifugal forces generated by the von Karman street from the trailing edge, particles were concentrated at its sides, weakening the signal in its center while strengthening it close to the suction side and at smaller radial coordinates. Anyway, except for the trailing edge region, uncertainties stay below 8% for both the absolute velocity and the flow angle.

$$\delta x = Za \cdot \frac{\sigma_x}{\sqrt{N_{records}}} \quad (1)$$

## PRESSURE PROBES RESULTS

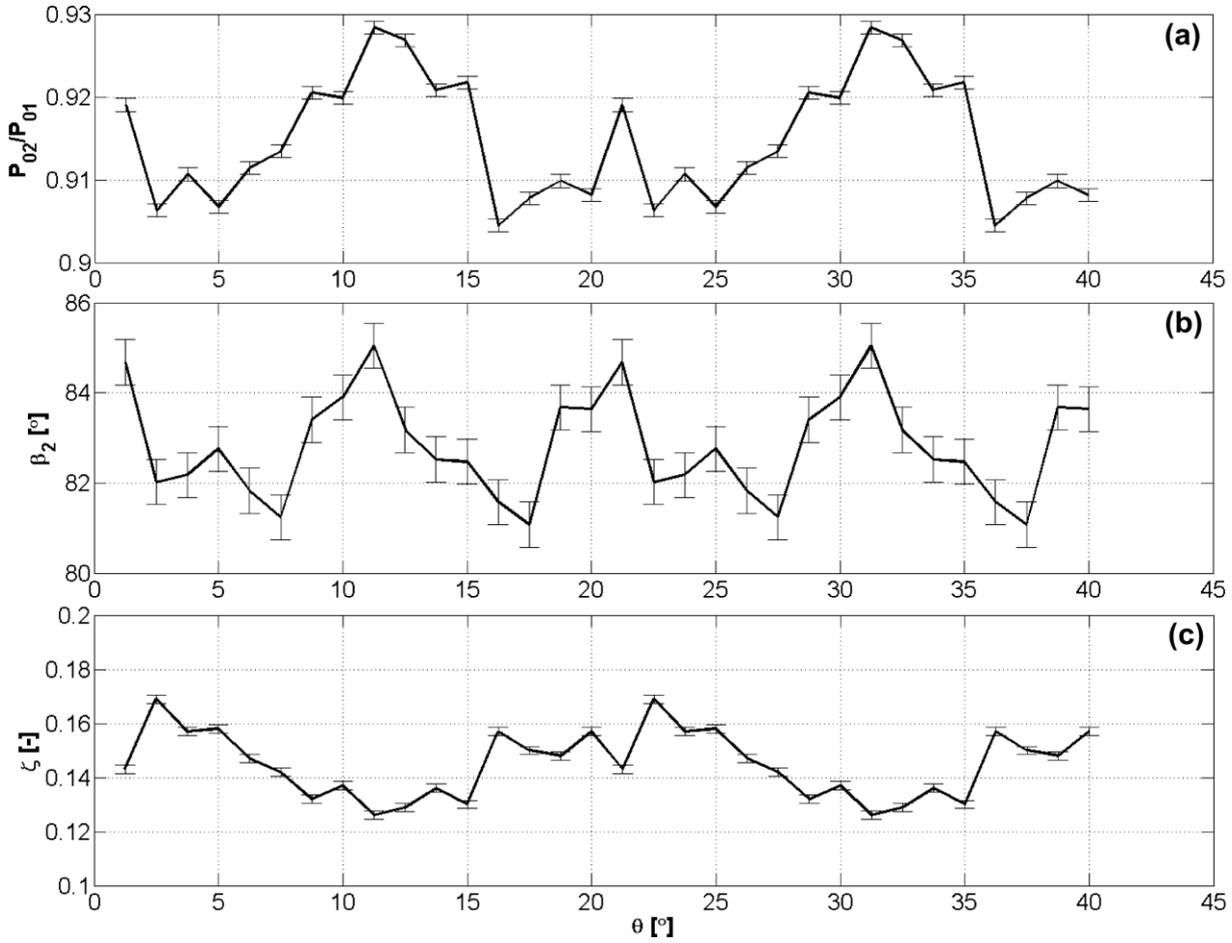
Figure 12a reports the sector total pressure wakes ( $P_{02}/P_{01}$ ) measured by means of the 16 micro virtual 3-hole probes. The retrieved distribution was cloned over two sectors to allow a clear interpretation. As suggested by the PIV campaign results, the extreme swirling of the flow had the tendency to turn the stream tubes towards the tangential direction, leading to a more compact radial extension and to an enhancement of the mixing-out of the flow field which, in turn, led to the inability to univocally distinguish between the “clean” and the “covered” wake. Anyway, a marked asymmetry can be recognized between the high total pressure regions centered respectively in  $\vartheta=11.25$  [°] and  $\vartheta=21.25$  [°], where the centers of the blade channels were most probably located. In general, peak values keep noticeably below 1, suggesting therefore a poor aerodynamic efficiency of the cascade which has to be entirely related to its very low aspect ratio.

The high pressure regions identified in the previous plot coincide with the peak values of the downstream flow angle. Its distribution is shown in Figure 12b and it is characterized by a much more pronounced periodicity with the wakes clearly located at  $\vartheta=7.5$  [°] and  $\vartheta=17.5$  [°] respectively. The flow angle magnitudes describe a highly tangential flow all over the sector, with a variability limited to less than 4 [°] between peaks and valleys.

The kinetic energy loss coefficient distribution is finally presented in Figure 12c as defined by Equation 2. It shows a definitely more homogeneous distribution over the sector with minima located in correspondence of the pressure peaks and of the high turning regions. The rather high loss values, always higher than 12%, are a further demonstration of how important is the effect of the aspect ratio on the global performance of the cascade.

$$\zeta = 1 - \frac{1 - \left(\frac{P_2}{P_{02}}\right)^{\frac{\gamma-1}{\gamma}}}{1 - \left(\frac{P_2}{P_{01}}\right)^{\frac{\gamma-1}{\gamma}}} \quad (2)$$

In order to reduce as much as possible the uncertainty on pressure measurements, small scale differential transducers were adopted and referenced to the upstream total pressure. The latter, instead, was acquired by a pressure scanner with respect to the atmospheric pressure. Such an arrangement of the measurement devices produced an uncertainty of about 26 [Pa] on probes readings and of 200 [Pa] for  $P_{01}$  which corresponds, in fact, to the highest contribution in the loss coefficient propagated error. In the case of the flow angle, the most important source of uncertainty was found to directly depend on the absolute probe positioning and it was estimated to be equal to 0.5 [°].



**Figure 12: micro virtual 3-hole probes results; (a) total pressure wakes, (b) flow angle taken from the radial direction, (c) kinetic energy loss coefficient.**

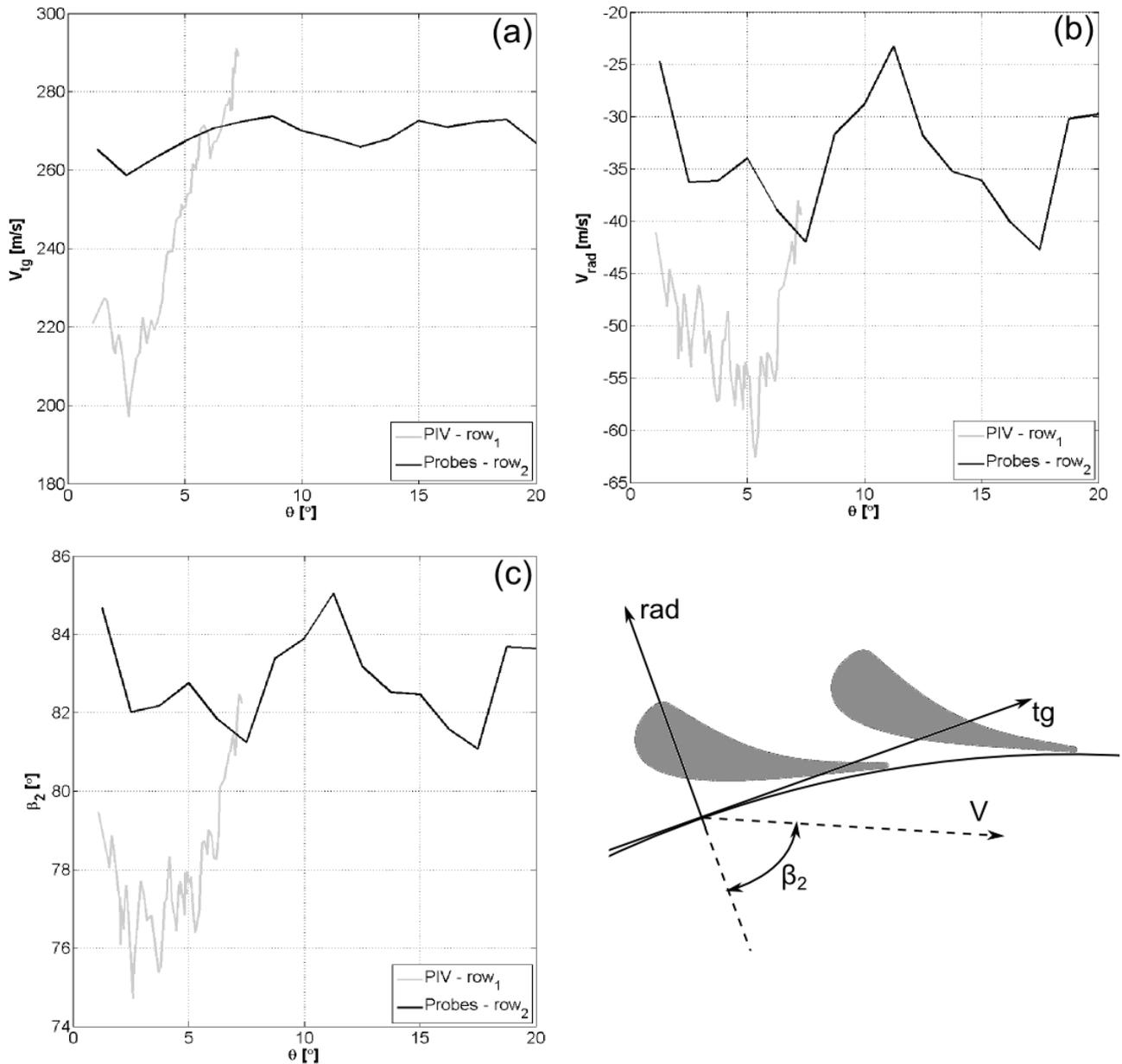
## PIV AND PROBES MEASUREMENTS COMPARISONS

A comparison is presented in terms of flow angle and velocity components between the pressure probes measurements and the PIV data extracted in correspondence of the first row of wall static pressure taps ( $8.6\%C_r$ ). Some assumptions had to be made in order to retrieve the velocity profiles from probes results, assumptions which are worth to critically revise.

As previously mentioned, pressure probes measurements were carried out exactly at the location of the second row of wall static pressure taps ( $44.5\%C_r$ ), which actually allowed to compute a local Mach number ( $Ma_2$ ) under the assumption of an homogeneous span-wise distribution of  $P_{2-row2}$ . Even though the latter hypothesis had to be necessarily accepted, it also corresponded to a very crude approximation of the problem: at the measurement location, in fact, an opening is present (see Figure 2) which increases the test section span of about 20%, naturally inducing important pressure gradients. Once the local Mach number was computed, a local temperature value had to be imposed to retrieve the velocity vector. In the present application, it was chosen to consider the test section as adiabatic and to impose therefore the downstream total temperature equal to  $T_{01}$ . This last assumption, much less drastic than the preceding one, was supported by the fact that the drop in total inlet temperature (typical of blow-down facilities) was small enough to lead to negligible heat transfer phenomena between the test section and the surrounding environment.

As previously commented in the PIV results discussion, the reduction of the radial coordinate at which the flow is flowing, necessarily implies an increase of the velocity level as an effect of the conservation of the angular momentum which, in turns, should lead therefore to a magnitude rising of both the velocity components. This is verified in fact for the tangential velocity  $V_{tg}$  (Figure 13a) which was globally boosted all over the sector span when moving from row 1 to row 2, with a very marked increase (about 25%) especially in the center of the wake.

On the contrary, as shown in Figure 13b, a 30% reduction of the radial velocity component ( $V_{rad}$ ) was retrieved between the two measurement points. This result, in contrast with what was stated before, has to be related to the presence of the test section opening which augmented the flow cross-section of about 20%. As a consequence,  $\beta_2$ , presented in Figure 13c, became more tangential when flowing further downstream, with a general increase of about 6 [°] all over the sector.



**Figure 13: comparison between PIV measurements extracted at 8.6% $C_r$  and probes measurements (44.5% $C_r$ ) in terms of tangential velocity component (a), radial velocity component (b) and flow angle (c). Conventions are reported in the bottom right.**

## CONCLUSIONS

An extensive measurement campaign was carried out at VKI with the intent of characterizing the aerodynamic performance of an ultra-low aspect ratio centripetal turbine stator. At the authors' best knowledge, this is the first time that such a design is experimentally investigated and described at this level of detail.

Two types of measurement techniques were used: PIV was carried out right downstream of the trailing edge circumference while micro cylindrical pressure probes were operated in virtual 3-hole

mode further downstream, where the test section was presenting an opening in the span-wise direction. The results describe a strongly swirling flow field whose tangential velocity was increasing when flowing towards smaller radii. The radial velocity component had exactly the opposite behavior: further downstream, in correspondence with the span-wise aperture,  $V_{rad}$  experienced a drastic 30% reduction to balance the bigger flow cross-section. Anyway, the absolute velocity globally grew towards the center of the centripetal cascade.

In general, this component showed a quite poor aerodynamic performance with kinetic energy loss levels as high as 16% which have to be completely related to the extremely small aspect ratio.

## ANKNOWLEDGEMENTS

The authors wish to express their gratitude to SNECMA (Groupe Safran) for the financial support to the measurement campaign and to Dr. Tamas Regert and to Professor Maria Rosaria Vetrano for their precious help in the building of the PIV setup.

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